

IN THE CLAIMS:

The following is a complete listing of all of the claims. Please amend the claims as follows:

1. **(Currently Amended)** A leading edge member for an aircraft, the aircraft having a substructure, at least a portion of the substructure being attached to a partial airfoil skin, the partial airfoil skin being a first airfoil portion member of an airfoil, the leading edge member comprising:

an exterior surface and an opposing interior surface forming a surface thickness therebetween;

wherein at least one pocket is recessed into the interior surface of the leading edge member, each pocket defining a region of the leading edge member having a pocket thickness that is less than the surface thickness of the leading edge member, each pocket being configured to deform in response to an impact from an object with the leading edge member, the at least one pocket being disposed solely within the leading edge member;

wherein the leading edge member is configured for attachment to the substructure, such that the exterior surface of the leading edge member forms a second airfoil ~~portion~~ member of ~~[[said]]~~ the airfoil.

2. **(Previously Presented)** The leading edge member according to claim 1, wherein the leading edge member forms the leading edge of a wing member.

3. **(Previously Presented)** The leading edge member according to claim 1, wherein the leading edge member forms the leading edge of a horizontal stabilizer.

4. **(Previously Presented)** The leading edge member according to claim 1, wherein the leading edge member forms the leading edge of a vertical fin.

5. **(Previously Presented)** The leading edge member according to claim 1, wherein the pockets are formed by a chemical etching process.

6. **(Previously Presented)** The leading edge member according to claim 1, wherein the pockets are formed by a mechanical milling process.

7. **(Previously Presented)** The leading edge member according to claim 1, wherein the leading edge member is curved about a longitudinal axis so as to form an upper airfoil surface and a lower airfoil surface.

8. **(Previously Presented)** The leading edge member according to claim 7, wherein the at least one pocket comprises:

a plurality of pockets arranged in a selected pattern over the interior surfaces of the upper airfoil surface and the lower airfoil surface.

9. **(Previously Presented)** The leading edge member according to claim 8, wherein each pocket is formed in one of the following geometric shapes: circle, oval, rectangle, square.

10. **(Previously Presented)** The leading edge member according to claim 8, wherein the pattern of pockets on the interior surface of the upper airfoil surface is a mirror image of the pattern of pockets on the interior surface of the lower airfoil surface.

11. **(Previously Presented)** The leading edge member according to claim 8, wherein the pattern of pockets on the interior surface of the upper airfoil surface is not a mirror image of the pattern of pockets on the interior surface of the lower airfoil surface.

12. **(Previously Presented)** The leading edge member according to claim 1, further comprising:

at least one rib member connected to the interior surface of the leading edge member for attaching the leading edge member to the substructure.

13. **(Previously Presented)** The leading edge member according to claim 1, further comprising:

a stiffening means connected to the interior surface of the leading edge member for providing localized stiffness to the leading edge member.

14. **(Previously Presented)** The leading edge member according to claim 13, wherein the stiffening means is an elongated I-shaped beam.

15. **(Previously Presented)** The leading edge member according to claim 13, wherein the stiffening means is not connected to the substructure.

16. **(Previously Presented)** The leading edge member according to claim 13, wherein the stiffening means is also connected to the substructure.

17. – 20. **(Cancelled).**

21. **(Previously Presented)** The leading edge member according to claim 1, wherein the leading edge member is attached to the substructure using at least one fastener.

22. **(Previously Presented)** The leading edge member according to claim 21, wherein the leading edge member is configured for detachment from the substructure by removing the at least one fastener.

23. **(Currently Amended)** The leading edge member according to claim 1, wherein the second portion member of said the airfoil is upstream from the first portion member of said the airfoil.

24. **(Previously Presented)** The leading edge member according to claim 1, wherein the leading edge member is upstream from the partial airfoil skin.